

GUJARAT TECHNOLOGICAL UNIVERSITY**BE - SEMESTER– VI (NEW) EXAMINATION – WINTER 2021****Subject Code:3160113****Date:02/12/2021****Subject Name:Advance Aerodynamics****Time:10:30 AM TO 01:00 PM****Total Marks: 70****Instructions:**

1. Attempt all questions.
2. Make suitable assumptions wherever necessary.
3. Figures to the right indicate full marks.
4. Simple and non-programmable scientific calculators are allowed.

| | | MARKS |
|------------|---|-----------|
| Q.1 | (a) Define Hypersonic flow in technical language. | 03 |
| | (b) What is the difference between aero thermodynamics and aerodynamics? | 04 |
| | (c) Explain construction of subsonic open type wind tunnel with neat sketch. | 07 |
| Q.2 | (a) Prove $C_p = 2 \sin^2 \theta$ for Newtonian theory. | 03 |
| | (b) What is the difference between supersonic and hypersonic wind tunnel? | 04 |
| | (c) Explain Solid blockage and wake blockage | 07 |
| | OR | |
| | (c) Explain Centrifugal force corrections to Newtonian theory with neat sketch. | 07 |
| Q.3 | (a) Define wind tunnel balances. | 03 |
| | (b) Explain Hypersonic expansion wave relation. | 04 |
| | (c) Explain Aerodynamic heating. | 07 |
| | OR | |
| Q.3 | (a) List out flow visualization techniques. | 03 |
| | (b) Define The velocity potential equation. | 04 |
| | (c) Explain Critical Mach number and Drag divergence mach number. | 07 |
| Q.4 | (a) What is The sound barrier and Area Rule. | 03 |
| | (b) With neat sketch explain thin shock layer. | 04 |
| | (c) Define Supercritical Airfoil, Rayleigh flow and Fanno flow. | 07 |
| | OR | |
| Q.4 | (a) Enlist Application to supersonic airfoils. | 03 |
| | (b) Explain Shock expansion method. | 04 |
| | (c) Define The velocity potential equation. | 07 |
| Q.5 | (a) Define “High temperature flow” characteristic for hypersonic flow. | 03 |
| | (b) Explain Low density flow. | 04 |
| | (c) Why car is not fly at Subsoinc, Supersoinc and hypersonic speed? | 07 |
| | OR | |
| Q.5 | (a) Explain flow over an airfoil case for hypersonic case. | 03 |
| | (b) Difference between supersonic flow and hypersonic flow. | 04 |
| | (c) Derive the Linearized supersonic pressure coefficient formula. | 07 |
